

## Exercises for Section 6.8, Exercise 3 (p. 271)

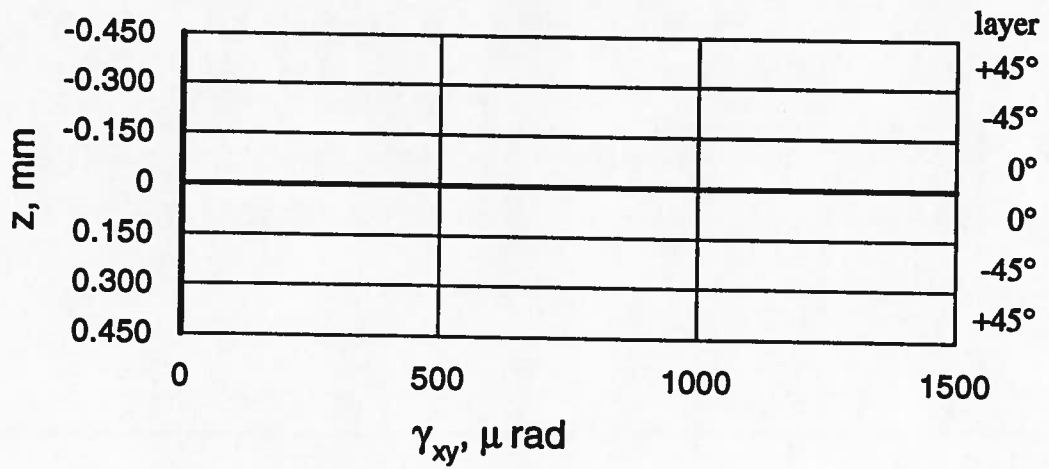
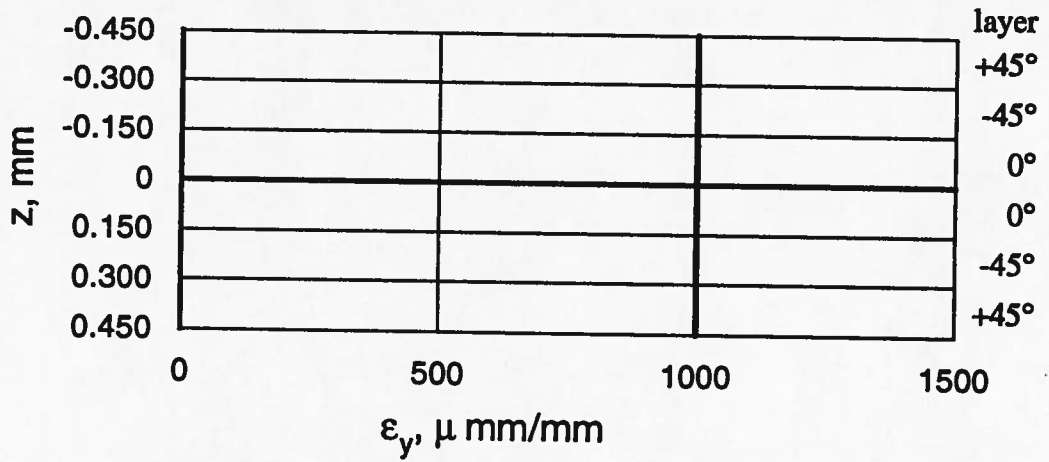
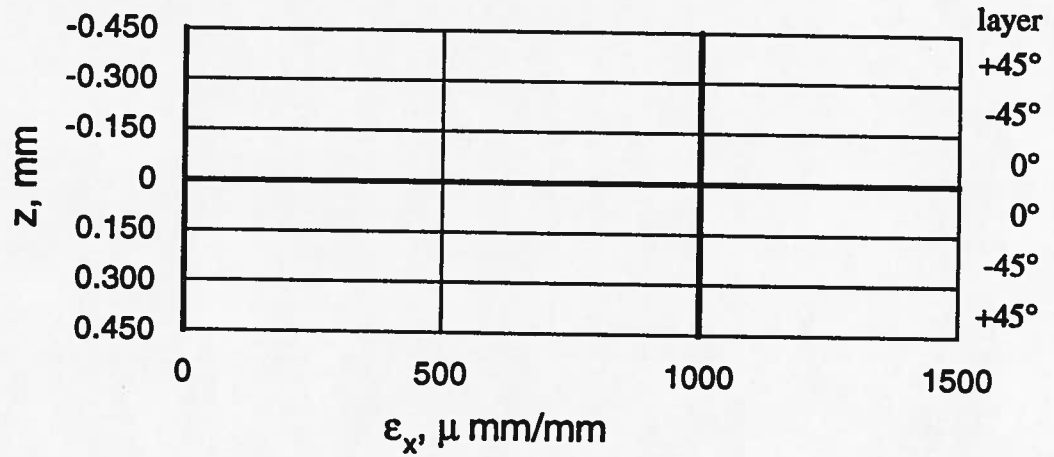
To solve this Exercise, the Computer Exercise was done first, as suggested. The solution to the problem will be presented in terms of the output from the suggested programming steps. In fact, the program used is the program that was developed over the course of the entire book, and is program *stress.f*, available to adopters of the book. That program is more fully developed than just the steps suggested in the current Computer Exercise. Presented will be a copy of the terminal session resulting from the interactive execution of the program, and a copy of the output from the program. Both have been cut and pasted somewhat to reflect the fact that all the theory in *stress.f* is not needed to solve this Exercise and has not been fully discussed yet. If you use *stress.f* it will ask for a temperature increment,  $\Delta T$ , and the output will contain information that will be presented in ch. 11 when free thermal strains are discussed further. Enter zero when asked to enter the temperature increment, and ignore the output lines related to free thermal strain effects. Also, the program prints out the *A*, *B*, and *D* matrices for the laminate. These will be discussed in ch. 7. Ignore that output for the time being. To follow is a brief description of the output for this biaxial stretching exercise. The output can be found on subsequent pages

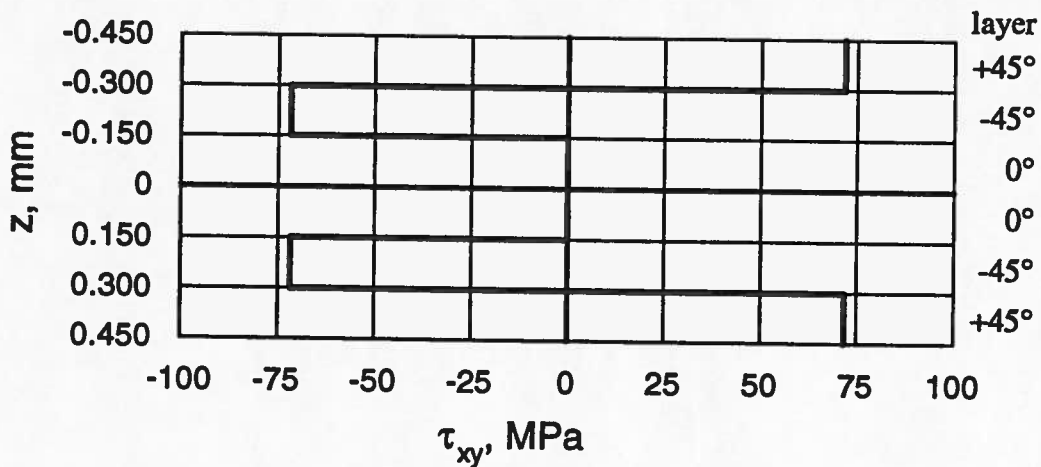
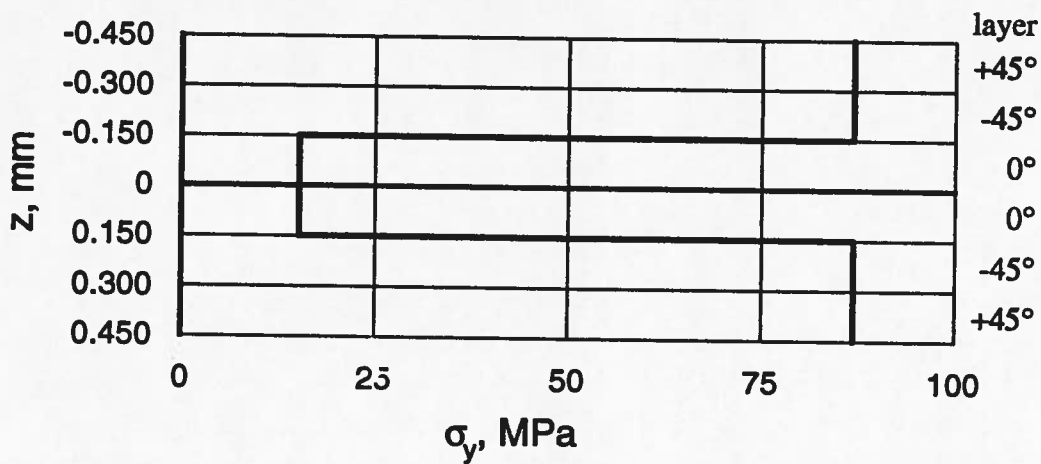
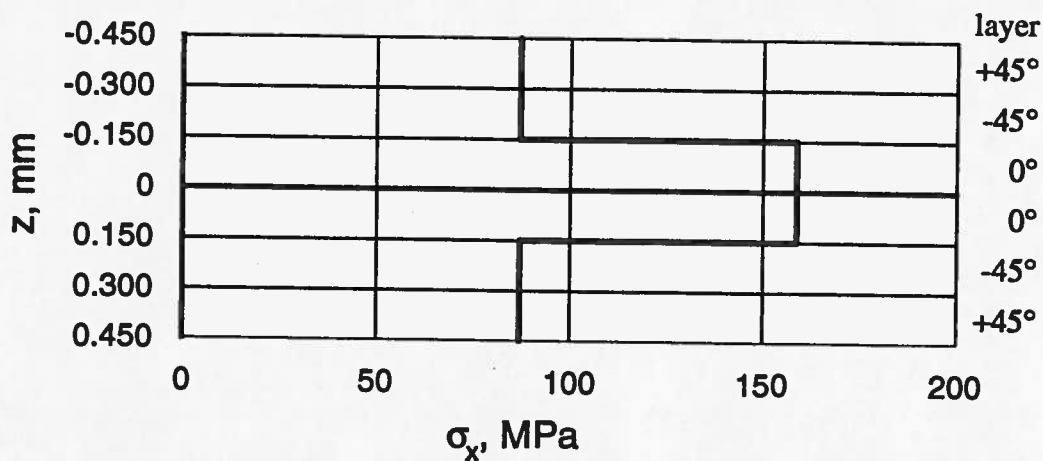
## Terminal Session

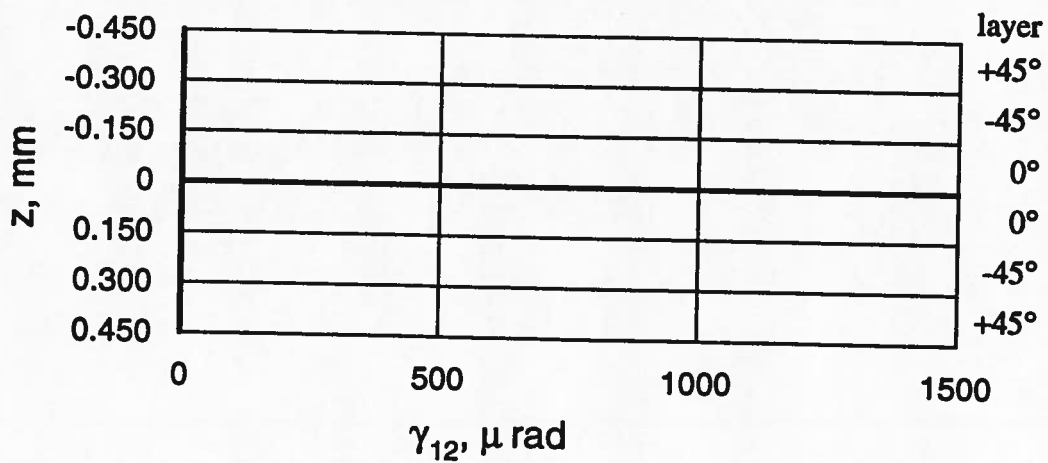
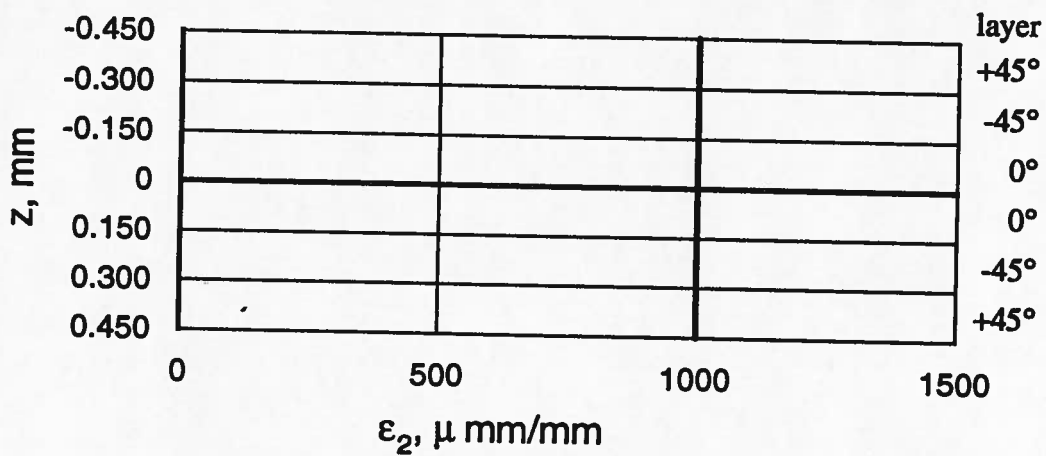
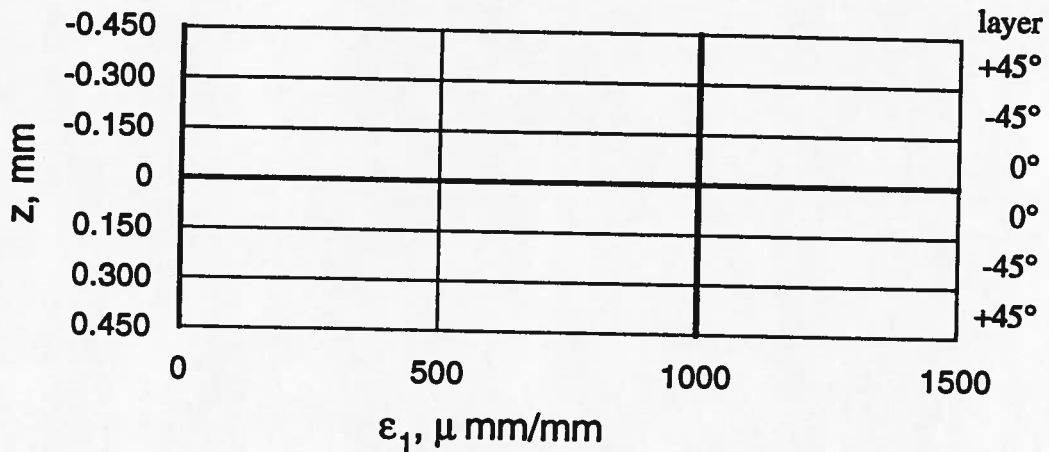
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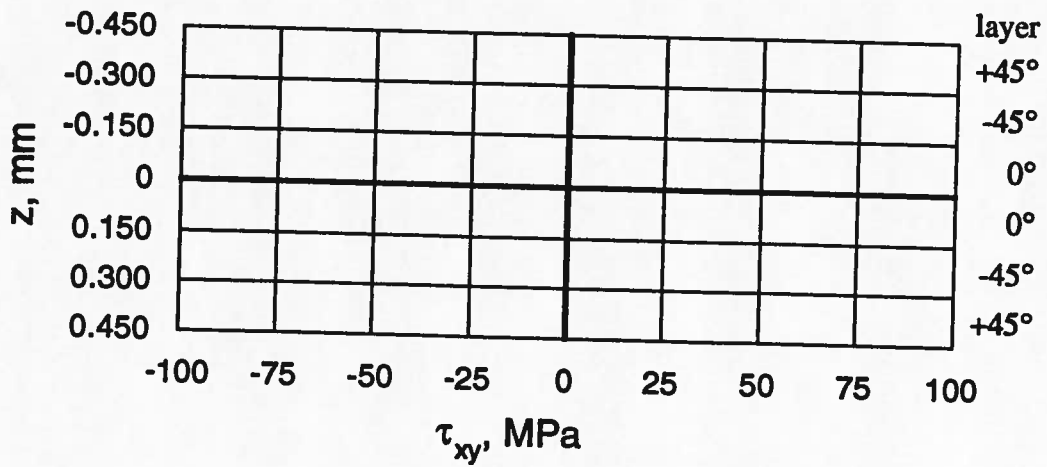
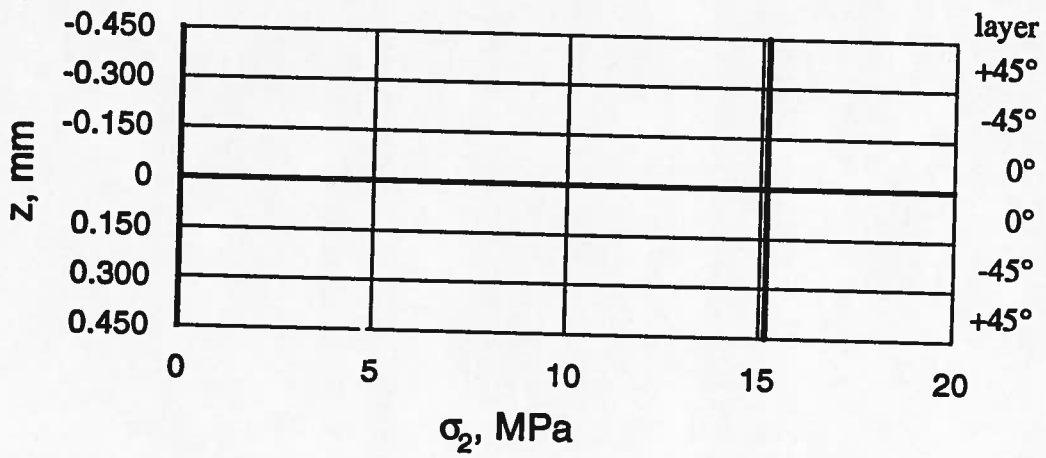
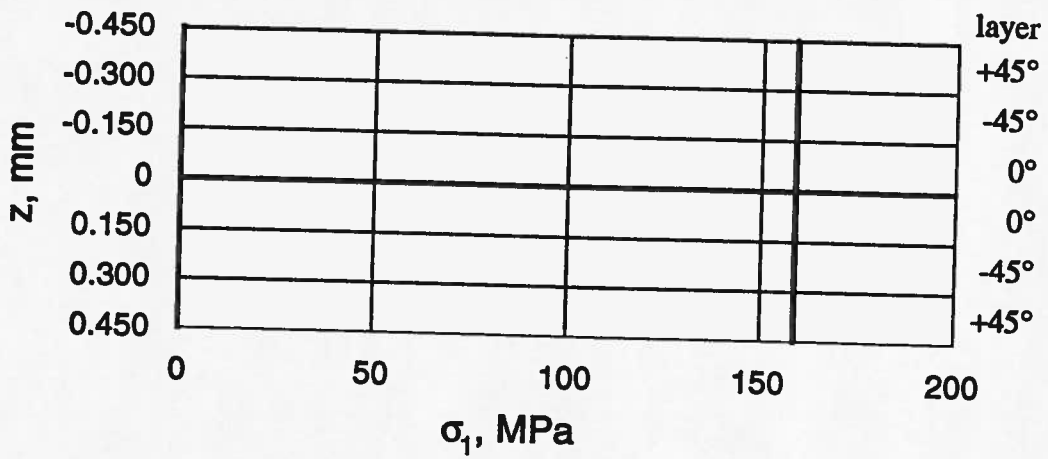
ENTER NUMBER OF LAYERS AND INTEGER KK
KK=0, ALL LAYERS SAME MATERIAL
KK=1, EACH LAYER A DIFFERENT MATERIAL
6 0
ENTER E11, E22, G12, NU12, AND THICKNESS
155.e9 12.1e9 4.4e9 0.248 150.e-6
ENTER ALPHA1 AND ALPHA2
-0.018e-6 24.3e-6
ENTER ORIENTATION (DEGREES) OF LAYER 1
45
ENTER ORIENTATION (DEGREES) OF LAYER 2
-45
ENTER ORIENTATION (DEGREES) OF LAYER 3
0
ENTER ORIENTATION (DEGREES) OF LAYER 4
0
ENTER ORIENTATION (DEGREES) OF LAYER 5
-45
ENTER ORIENTATION (DEGREES) OF LAYER 6
45
ENTER TEMPERATURE INCREMENT
0
RESULTANTS OR REF SURFACE STRAINS AND CURVATURES?
1=RESULTANTS, 2=STRAINS AND CURVATURES
2
ENTER STRAINS AND CURVATURES
EPSX, EPSY, GAMXY, KX, KY, KXY
1000.e-6 1000.e-6 0 0 0 0
[Process completed]

```









## Exercises for Section 6.8, Exercise 4 (p. 272)

(Note: In part (c) of the Exercise, the comment regarding  $M_{xy}$  is incorrect. For this problem  $M_{xy}$  is identically zero.)

Below is the output cut and pasted from program *stress.f*. Following that are the answers to the questions in the Exercise, and the plots of the strains and stresses. This problem has unusual distributions of strains and stresses.

LAYER NO. 1 THETA = 45.0 THICKNESS = 0.15000E-03  
 E1 = 0.1550E+12 G12 = 0.4400E+10 ALPHA1 = -0.1800E-07  
 E2 = 0.1210E+11 NU12 = .248 ALPHA2 = 0.24300E-04

Z( 0)=-0.45000E-03 Z( 1)=-0.30000E-03 (SEE FIG. 6.3)

Q MATRIX, EQ. 4.17			QBAR MATRIX, EQ. 5.84		
0.1557E+12	0.3015E+10	0.0000E+00	0.4788E+11	0.3908E+11	0.3590E+11
0.3015E+10	0.1216E+11	0.0000E+00	0.3908E+11	0.4788E+11	0.3590E+11
0.0000E+00	0.0000E+00	0.4400E+10	0.3590E+11	0.3590E+11	0.4047E+11

COEFFICIENTS OF THERMAL EXPANSION, EQ. 5.143

ALPHAX = 0.1214E-04 ALPHAY = 0.1214E-04 ALPHAXY = -0.2432E-04

\* \* \* \* \*

LAYER NO. 2 THETA = -45.0 THICKNESS = 0.15000E-03

E1 = 0.1550E+12 G12 = 0.4400E+10 ALPHA1 = -0.1800E-07  
 E2 = 0.1210E+11 NU12 = .248 ALPHA2 = 0.24300E-04

Z( 1)=-0.30000E-03 Z( 2)=-0.15000E-03 (SEE FIG. 6.3)

Q MATRIX, EQ. 4.17			QBAR MATRIX, EQ. 5.84		
0.1557E+12	0.3015E+10	0.0000E+00	0.4788E+11	0.3908E+11	-0.3590E+11
0.3015E+10	0.1216E+11	0.0000E+00	0.3908E+11	0.4788E+11	-0.3590E+11
0.0000E+00	0.0000E+00	0.4400E+10	-0.3590E+11	-0.3590E+11	0.4047E+11

COEFFICIENTS OF THERMAL EXPANSION, EQ. 5.143

ALPHAX = 0.1214E-04 ALPHAY = 0.1214E-04 ALPHAXY = 0.2432E-04

\* \* \* \* \*

LAYER NO. 3 THETA = .0 THICKNESS = 0.15000E-03

E1 = 0.1550E+12 G12 = 0.4400E+10 ALPHA1 = -0.1800E-07  
 E2 = 0.1210E+11 NU12 = .248 ALPHA2 = 0.24300E-04

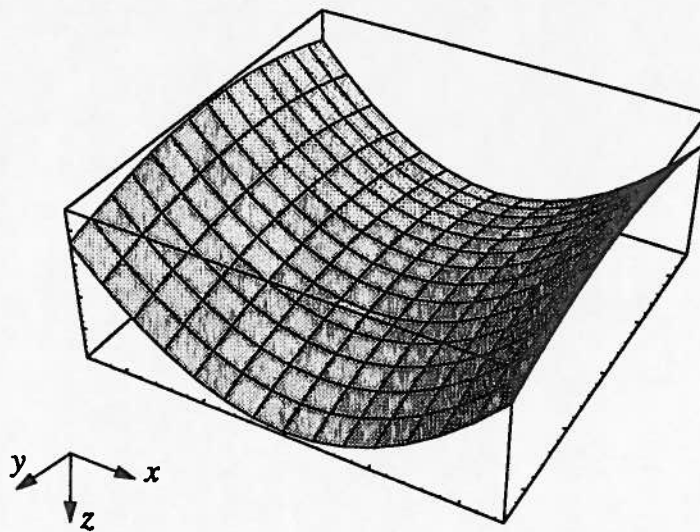
Z( 2)=-0.15000E-03 Z( 3)= 0.29104E-10 (SEE FIG. 6.3)

## COMPUTED STRESS RESULTANTS

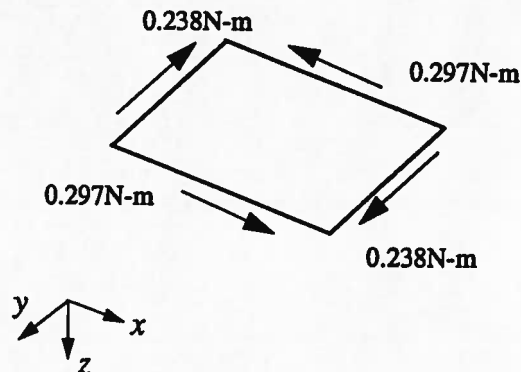
$N_{xM} = 0.00000E+00$   $N_{yM} = 0.00000E+00$   $N_{xYM} = 0.00000E+00$   
 $M_{xM} = 0.19077E+01$   $M_{yM} = -0.11897E+01$   $M_{xYM} = 0.00000E+00$

The magnitude of moment  $M_x$  is greater than the magnitude of moment  $M_y$  because of the presence of the  $0^\circ$  layers. Though the  $0^\circ$  layers are near the reference surface, and thus have minimal influence on bending response, they do have an influence, offering some resistance to bending because of the stiff fibers.

Deformed shape



Required edge moments (all N's = 0)

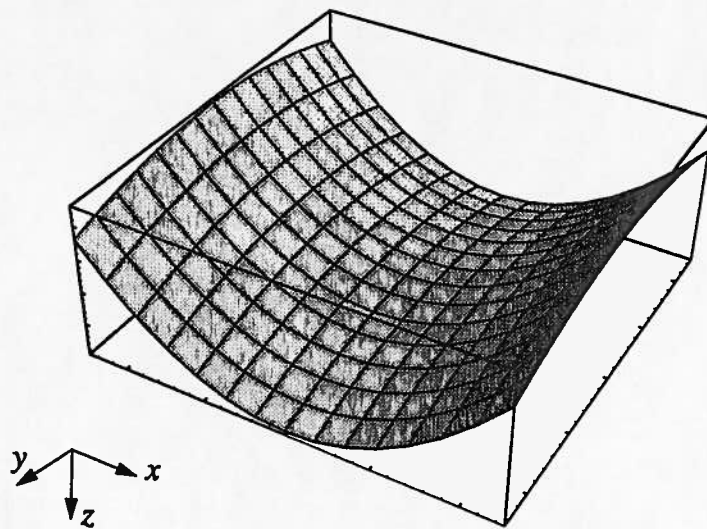


## COMPUTED STRESS RESULTANTS

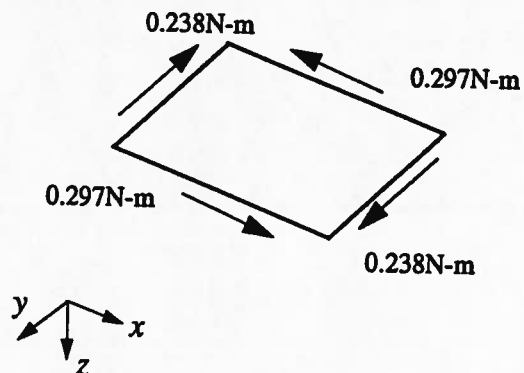
$N_{xM} = 0.00000E+00$   $N_{yM} = 0.00000E+00$   $N_{xYM} = 0.00000E+00$   
 $M_{xM} = 0.19077E+01$   $M_{yM} = -0.11897E+01$   $M_{xYM} = 0.00000E+00$

The magnitude of moment  $M_x$  is greater than the magnitude of moment  $M_y$  because of the presence of the  $0^\circ$  layers. Though the  $0^\circ$  layers are near the reference surface, and thus have minimal influence on bending response, they do have an influence, offering some resistance to bending because of the stiff fibers.

Deformed shape



Required edge moments (all N's = 0)



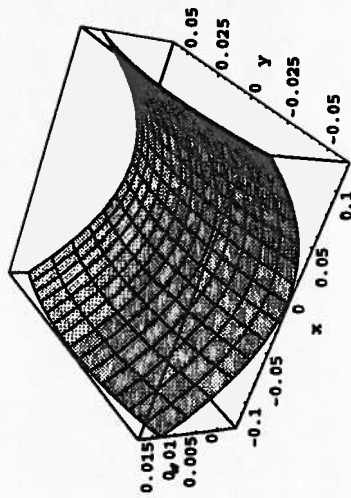


Ex6.8\_4.mn

```

In[2]:=
kx = 2.22222
ky = -2.22222
Out[1]=
2.22222
Out[2]=
-2.22222
In[3]:=
w[x_,y_] := (kx*x + ky*y)/2
In[4]:=
Plot3D[w[x,y], {x,-0.125,0.125}, {y,-0.0625,0.0625},
  AxesLabel-> {x,y,w}]

```



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Out[4]=
-SurfaceGraphics-

```

